

CHECK WIRING HARNESS BEFORE INSTALLATION

It's easier/cheaper to check and recheck all harness's and connections than repairing them later!

TAS 1000

INSTALLATION PROCEDURES

DOCUMENT 1000-402

DATE: DEC 10, 2010 REV.H

PLEASE READ INSTRUCTIONS
COMPLETELY
BEFORE PROCEEDING
WITH INSTALLATION

Insight

Instrument Corp.

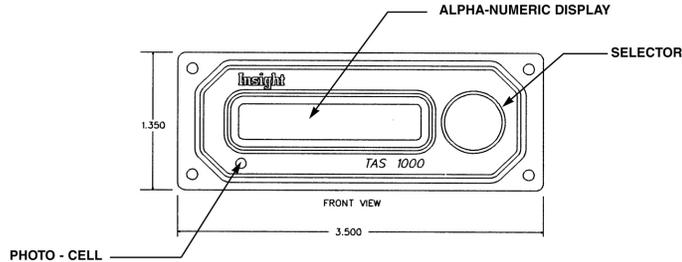
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PLANNING THE INSTALLATION

The TAS 1000 Air Data System interconnects with several existing aircraft systems. Full implementation requires connection with: Electrical system, Pitot/Static system, OAT probe, slaved heading system, fuel flow sensors, altimeter, and GPS.



Note that the TAS 1000 contains an ambient light sensor for automatic brightness control and so requires no connection to aircraft lighting bus. Read all installation instructions and examine the aircraft to determine existing aircraft equipment.

1.1 TAS SYSTEM INSTALLATION OPTIONS

OPTIONS	REQUIREMENTS	RESULTS
FUEL FLOW		
A. INSIGHTS FUEL FLOW SENSOR	CONNECT TAS TO INSIGHT FUEL FLOW SENSOR	TAS WILL PROVIDE FUEL DATA
B. AC SINEWAVE FUEL FLOW SENSOR	CONNECT TAS TO EXISTING AC SINEWAVE FUEL FLOW SENSOR AND INSTRUMENT	TAS WILL PROVIDE FUEL DATA AS BACK-UP TO EXISTING INSTRUMENT
C. FLOWSCAN 201B AND EXISTING INSTRUMENT	CONNECT TAS TO EXISTING 201B FUEL FLOW SENSOR AND INSTRUMENT	TAS WILL PROVIDE FUEL DATA AS BACK-UP TO EXISTING INSTRUMENT
D. FLOWSCAN 201B ONLY	CONNECT TAS TO EXISTING 201B FUEL FLOW SENSOR ONLY	TAS WILL PROVIDE FUEL DATA
E. TF 500 FUEL SENSOR	CONNECT TF500 FUEL INPUTS AS SHOWN	TF500 WILL PROVIDE FUEL FUNCTIONS TO A GPS
F. NONE	CONNECT TAS FUEL FLOW INPUTS TO GROUND	NO FUEL FUNCTIONS AVAILABLE
COMPASS		
A. NO COMPASS SYSTEM	IF NO HEADING SOURCE IS AVAILABLE CONNECT TAS AS SHOWN	PILOT MUST ENTER HEADING CHANGES ON TAS
B. SYNCHRO (SLAVED DG/HSI)	WIRE TAS SYNCHRO INPUTS TO AIRCRAFT X, Y, Z AND 400 HZ REF.	TAS WILL UTILIZE HEADING FROM COMPASS.
C STEPPER (KING HSI)	CONNECT TAS SA, SC INPUTS TO HSI OUTPUTS AS SHOWN.	AS ABOVE, BUT PILOT WILL PERIODICALLY CHECK AND RESET TAS HEADING.
BARO SETTINGS		
A. BARO-OUTPUT ALTIMETER	CONNECT ALTIMETER BARO-OUTPUT TO TAS INPUT PINS. (400 HZ REF. REQUIRED)	TAS WILL ACQUIRE BARO-SETTING FROM THE ALTIMETER
B. NONE	CONNECT TAS BARO INPUT PINS TO GROUND	PILOT MUST SET BARO ON TAS

2. HARDWARE INSTALLATION

2.0 INSTALL TAS WIRING

Install all wiring in accordance with AC 43.13-1B Chapter 11 Section 8, 11-96, 11-97, 11-98, 11-100, 11-104, 11-105, 11-106, and 11-107.

2.1 INSTALLING THE TAS 1000 AIR DATA COMPUTER

Select a suitable panel location for the TAS 1000. Avoid locations subject to direct sunlight. Check for accessibility to the pitot/static and all other required systems. Use existing or create a panel hole in accordance with Insight Drawing #1000-304 (PANEL CUT-OUT).

Mount TAS 1000 using four supplied screws (or similar).

2.2 INSTALLING THE WINDICATOR

The Insight Windicator displays the TAS 1000's real-time wind aloft data in graphical format. It may be mounted in any convenient location. Mount the Windicator in accordance with Insight Drawing #1000-305 (WINDICATOR MOUNTING DIAGRAM). Terminate in accordance with Insight Drawing #1000-301 Page 1 (TAS 1000 INSTALLATION SCHEMATIC).

2.3 CONNECTING POWER AND GROUND

Using Insight supplied connector and installer supplied circuit-breaker connect to aircraft avionics bus in accordance with Insight Drawing #1000-301 Page 1.

2.4 INSTALLING THE OAT (Outside Air Temperature) SENSOR

Select a suitable location on the exterior of the aircraft for the OAT sensor. Choose a location on the belly of the aircraft avoiding vents, exhaust, or any feature that may interfere with sensing ambient air temperature. Mount the sensor in accordance with Insight Drawing #1000-302 (OAT MOUNTING DIAGRAM). Terminate the OAT cable in accordance with Insight Drawing #1000-301 Page 1.

2.5 INSTALLING THE EXPANISON MODULE (Altitude Encoder & Over-speed Indicator)

Select a suitable location for the EXPANISON MODULE. Choose a location inside the aircraft avoiding radio transmitters, antennas or any feature that may induce RF noise within the EXPANISON MODULE. Avoid installation in places that may become extremely hot during operation. Mount the sensor in accordance with Insight Drawing #1000-302 (EXPANISON MODULE MOUNTING DIAGRAM). Terminate the EXPANISON MODULE cable in accordance with Insight Drawing #1000-301 Page 4.

2.6 INSTALLING THE FUEL FLOW SENSOR(S)

Aircraft without existing FF sensors should have one Insight supplied sensor installed per engine. Existing compatible FF sensors may be connected directly to the TAS 1000. Install new sensors in accordance with Insight Drawing #1000-303 (FUEL FLOW/TF SENSOR MOUNTING DIAGRAM). Use Insight Drawing #1000-301 page 2 to cable and terminate Fuel Flow Sensor #1000-004. Use Insight Drawing #1000-816 to cable and terminate TF500 Sensor #1000-006. Refer to AC 43.13-1B, Chapter 8 Section 2, 8-30, 8-31, 8-37, 8-38. **Fuel flow case must be connected to airframe ground.**

NOTE: TF500 Sensor needs to be connected to a GPS and can operate independently from TAS 1000 system.

2.7 CONNECTING TO A HEADING SYSTEM (synchro system preferred)

TAS 1000 should be connected to either a slaved synchro compass system OR a digital stepper heading system (KCS-55). Autopilot outputs are usually not compatible. Aircraft not equipped with slaved heading systems will require the pilot to manually input heading data to the TAS 1000. Refer to Insight Drawing #1000-301 Page 1 for termination.

2.8 CONNECTING TO AN ALTIMETER (Baro setting)

TAS 1000 may be connected to an altimeter to automatically receive barometer-correction (altimeter setting). Aircraft not equipped with barometer-setting output altimeters will require the pilot to manually input altimeter setting to the TAS 1000. Terminate in accordance with Insight Drawing #1000-301 Page 1. Note that if connection is made to an altimeter, the TAS 1000 barometer setting MUST be calibrated using the TAS 1000 DEALER MODE.

2.9 CONNECTING TO A GPS

TAS 1000 should be connected to a GPS. See the DEALER MODE section for compatible GPS models. Cable and terminate in accordance with Insight Drawing #1000-301 Page 1. Note that the serial protocol configuration is required using the TAS 1000 DEALER MODE.

3.0 CONNECTING TO A PITOT/STATIC SYSTEM

Connect the TAS 1000 to a PITOT/STATIC system using the Insight supplied air fitting or equivalent. Installer must supply approved tubing. Refer to AC 43.13-1B Chapter 12 Sections 1, 12-1, 12-2, Section 2, 12-8, 12-9, 12-19, 12-20, Section 4, 12-51.

Make all connections to provide for moisture drainage. Aircraft equipped with Primary-Secondary Pitot/Static systems should have the TAS 1000 connected to the Secondary system. Perform a leak check prior to flight tests.

4.0 TAS DEALER MODE SET-UP

DEALER MODE allows access to the following configuration items:

Single/Twin Engine

Fuel Units - Pounds/Gallons/Liters

Aviation gasoline - yes or no

Synchro – Synchro or HSI input

Baro 31 - Baro-pot input calibration 31 inches of Hg

Baro 28 - Baro-pot input calibration 28 inches of Hg

M.F.O. - Maximum fuel on board (gallons)

KL - Single or left engine (pulses/gallon AC mode)

KR - Right engine K factor (pulses/gallon AC mode)

A.W.B. - Altitude Alert warning band (feet)

RSOUT - RS - 232 output mode C, D, F, Z or G

EXIT

Start the TAS 1000 in DEALER MODE by pushing and holding the selector while applying power. The TAS 1000 will initially show "CONFIG", continue holding the selector for approximately 20 seconds or until the TAS 1000 shows "DEALER", then release selector.

4.1 Single/Twin Engine

The TAS 1000 default will show “SINGLE N”. For twin-engine aircraft push the selector to accept the default and advance the display to the next item. For single engine aircraft, turn the selector to show “SINGLE Y”. Push the selector to accept this selection, and advance the display to the next configuration item.

4.2 Fuel Units

The TAS 1000 allows you to select between gallons or pounds. To select a measurement unit, turn the selector to show “FUEL GAL OR FUEL LB”. Push to accept and proceed to next item.

4.3 Fuel Type (aviation gasoline or jet fuel)

default NO

For piston engine(s), select yes.

For turboprop and jet engine(s), select no.

4.4 SYNCHRO (YES or NO)

default YES

If connected to a synchro compass select yes.

For all others select no.

4.5 Barometer Setting Calibration 31 inches of Hg

default NO

The TAS 1000 Barometer Setting (Altitude Setting) must be calibrated if the TAS 1000 is connected to an altimeter’s Barometer Setting output.

To calibrate, set altimeter to 31.00 inches Hg and turn the TAS 1000 selector to show “BARO31 Y”. Push the selector to accept the setting. Failure to obtain a “BARO31Y” display indicates an invalid barometer signal.

Push the selector twice to omit this feature and proceed to next item.

4.6 Barometer Setting Calibration 28 inches of Hg

default NO

Set the altimeter to 28.00 inches Hg. The TAS 1000 shows “BARO28 N”. Turn the TAS 1000 selector to display “BARO28 Y” and push the selector to accept the setting. Failure to obtain a “BARO28 Y” display indicates an invalid Barometer signal.

4.7 Maximum Fuel on Board MFO

default 50 gallons

Dealer enters the maximum fuel on board available for that model of aircraft (gallons).

4.8 Fuel Flow Calibration (KL Left/Single Engine)

default 30000 pulses/gallon (AC mode)

For AC mode:

The TAS 1000 must be programmed with the correct K value for each connected fuel flow sensor. Fuel flow sensors differ in their sensitivity (K value expressed in Pulses per Gallon). The TAS 1000 can operate with K values ranging from 1000 to 99900. Insight supplied sensors come with nominal K values of 30000 (leave K value at 30000 for Insight supplied sensors). Sensors from other suppliers do not.

Turn the TAS 1000 selector to show the correct K value for the installed sensor on engine one, and push the selector to accept the value and proceed to the next item.

4.9 Fuel Flow Calibration (KR for Right Engine)

default 30000 pulses/gallon (AC mode)

For AC mode:

The TAS 1000 must be programmed with the correct K value for each connected fuel flow sensor. Fuel flow sensors differ in their sensitivity (K value expressed in Pulses per Gallon). The TAS 1000 can operate with K values ranging from 1000 to 99900. Insight supplied sensors come with nominal K values of 30000 (leave K value at 30000 for Insight supplied sensors). Sensors from other suppliers do not.

Turn the TAS 1000 selector to show the correct K value for the installed sensor on this engine, and push the selector to accept the value and proceed to the next item.

4.10 Altitude Warning Band AWB

default 300 feet

Enter altitude-warning band setting for altitude alert function.

4.11 RS-232 TRANSMIT FORMAT

default C

Turn the selector knob to format D, F, G or Z as appropriate for the connected GPS.

The TAS 1000 is compatible with the following GPS types;

GPS MANUFACTURER	MODEL	TAS TRANSMIT FORMAT
BENDIX/KING	KLN89/B	C OR D
BENDIX/KING	KLN90/B	C OR D
GARMIN	400/500 SERIES	G OR Z
APOLLO	GX SERIES	Z

TAS RS-232 TRANSMIT FORMATS:

DATA TYPE	C	D	F	G	Z
TOTAL FUEL REMAINING	YES	YES	YES	YES	NO
TOTAL FF RATE	YES	YES	YES	NO	NO
ENGINE 1 FF RATE	YES	YES	YES	YES	YES
ENGINE 2 FF RATE	YES	YES	YES	YES	YES
TOTAL FUEL USED	YES	YES	YES	NO	NO
ENGINE 1 FUEL USED	YES	YES	YES	YES	YES
ENGINE 2 FUEL USED	YES	YES	NO	YES	YES
IAS KNOTS	YES	YES	NO	YES	YES
TAS KNOTS	YES	YES	NO	YES	YES
MACH NUMBER	YES	YES	NO	YES	YES
PRESSURE ALTITUDE (FT)	YES	YES	NO	YES	YES
DENSITY ALTITUDE (FT)	YES	YES	NO	YES	YES
TOTAL AIR TEMPERATURE (°C)	YES	YES	NO	YES	YES
WIND DIRECTION (°M)	YES	YES	NO	YES	YES
WIND SPEED (KNOTS)	YES	YES	NO	YES	YES
DRIFT (DEGREES)	YES	YES	NO	YES	NO
RATE CHANGE OF ALTITUDE (FT / MIN)	YES	YES	NO	YES	YES
HEADING (°M)	YES	YES	NO	YES	YES
STATIC AIR TEMPERATURE (°C)	YES	YES	NO	YES	YES
BARO-SETTING (INCHES Hg)	NO	YES	NO	YES	NO
BARO-CORRECTED ALTITUDE	NO	YES	NO	YES	NO
FUEL MEASUREMENT UNITS	YES	YES	YES	NO	NO

NOTE: TAS RS-232 RECEIVES “AVIATION FORMAT” FOR FORMAT

NOTE: For GPS types not listed contact Insight Technical Support.

Turn the TAS 1000 selector to indicate the required code and push the selector to accept.
GPS must also be set-up to communicate with TAS. (See sec. 4)

See GPS manual.

4.12 EXIT DEALER MODE

Push the TAS 1000 selector to switch from DEALER MODE to NORMAL MODE.

5.0 INSTRUCTIONS FOR GARMIN 400/500 SERIES SOFTWARE SET UP:

- 1) To start set up mode.
 Press and hold “ENT” button and apply power.
 Hold “ENT” until “OK” message appears.
 Release “ENT” button.
 Momentary press “ENT” button once.
- 2) Turn inner rotary knob until “Main RS-232 Config” appears.
- 3) Use cursor and “ENT” button to select:

	INPUT	OUTPUT
Channel 1	Shadin – fadc	Avtn no alt

- 4) Turn power off.

5.1 FINAL ACCEPTANCE CHECK LIST

All aircraft systems connected to the TAS 1000 must also be checked for correct operation. Refer to AC 43.13-1B Chapter 12 Sections 1, 2, 3, 4 - 12-1, 12-8, 12-9, 12-17, 12-19, 12-20, 12-37.

The Pitot/Static system should be checked for leaks prior to flight test. Refer to the TAS 1000 Operating Instructions and conduct a ground check and a test flight before returning the aircraft to service.

GROUND CHECK

- 1. POWER-ON TAS.
- 2. WINDICATOR ALL-LED CHECK / WALKING DOT MODE.
- 3. TAS INITIAL DATA ENTRY (BR / HEADING / FUEL)
- 4. CHECK TAS AUTO-BRIGHT / DIM FUNCTION.
- 5. CROSS-CHECK EACH DATA ITEM ON THE TAS CONFIG LIST AGAINST ANOTHER DATA SOURCE.

NOTE: WINDICATOR DATA ONLY AVAILABLE WITH AIRCRAFT IN FLIGHT.

5.2 FLIGHT CHECK

- ONLY AFTER SUCCESSFULLY COMPLETING GROUND CHECK.**
- REPEAT ALL GROUND DATA INDICATIONS.**
- TAS FUEL FLOW INDICATIONS.**
- TAS ALTITUDE.**
- TAS VERTICAL SPEED.**
- WINDICATOR.**
- TAS WINDS ALOFT DATA.**

5.3 TAS SERVICE

1. The TAS-1000 system should have a long service life and can be used indefinitely on-condition. Insight recommends periodic testing and calibration of the TAS-1000 system every 2 years to ensure continued accuracy and precision.
2. In the unlikely event of operational difficulties during flight, flight crew should turn-off the TAS-1000 system by pulling the circuit-breaker, and report the incident to maintenance personnel. Refrain from further use of the TAS-1000 system in-flight until any service issues are resolved.
3. The TAS-1000 is dependent on internal and external sensors for data measurements and all resulting calculations. In the event that a data type is not available, the TAS-1000 display will show "N/A" or "-----".
4. The core of the TAS-1000 is the pitot/static measurement system with internal, temperature-controlled sensors. Air data will achieve best accuracy after the sensors stabilize at their design operating temperature of 45 degrees C which may require up to ten minutes after power-up. Select the "OV" data type with the selector knob to display the sensor temperature. The TAS-1000 sensors should maintain 45 +/- 0.1 degrees.
5. Data is only available from the TAS-1000 when data dependencies are satisfied. Refer to 5.4: TAS-1000 Data Sources to determine availability of data types. Note that some data types are highly dependent on other data. For example, wind calculations require True Airspeed, Heading, Groundspeed and Track and so are not available unless all four data types are available which cannot occur until the aircraft is in-flight.
6. No field repairs are allowed on the TAS-1000 system or components. Any malfunctioning TAS-1000 system or components must be removed from the aircraft and returned to Insight Instrument Corporation for repair. Removal can be accomplished by reversing the instructions for TAS-1000 component installation. The TAS-1000 wiring and connectors may remain in the aircraft while components are removed for service. Pitot tubes must be securely plugged while the p/n 1000-001 display is disconnected, and a pitot/static leak test performed on the aircraft before flight.

5.4: TAS-1000 Data Sources

Display	Data Type	Units	Dealer Adjust	Data Source(s)	Dependencies
AL	Pressure Altitude	Feet	N/A	Measurement of Static+BR	Internal Static Sensor+BR
BR	Barometric Setting	Inches of Mercury	Requires Cal.	Altimeter or Pilot Input	Baro-pot Altimeter
BR	Barometric Setting	millibars	Requires Cal.	Altimeter or Pilot Input	Baro-pot Altimeter
DA	Density Altitude	Feet	N/A	Calculated	Internal Static Sensor+BR+SAT
D/D	Density Altitude Difference	Feet	N/A	Calculated	DA+AL
IAS	Indicated Air Speed	Knots	N/A	Calculated	TAS+DA
IAS	Indicated Air Speed	Miles per Hour	N/A	Calculated	TAS+DA
TAS	True Air Speed	Knots	N/A	Calculated	MACH+SAT
TAS	True Air Speed	Miles per Hour	N/A	Calculated	MACH+SAT
GRS	Ground Speed	Knots	RSOUT	GPS Interface	Data only valid in-flight
GRS	Ground Speed	Miles per Hour	RSOUT	GPS Interface	Data only valid in-flight
TRK	Track	Degrees Magnetic	RSOUT	GPS Interface	Data only valid in-flight
MACH	Mach Number	Fraction of Mach One	N/A	Measurement+Calculation	Internal Static+Dynamic Sensors
VS	Vertical Speed	Feet per Minute	N/A	Calculated	AL
GR	Gradient	Feet per Nautical Mile	N/A	Calculated	AL+TRK
WSP	Wind Speed	Knots	N/A	Calculated	TAS+HDG+GS+TRK
WSP	Wind Speed	Miles per Hour	N/A	Calculated	TAS+HDG+GS+TRK
WDR	Wind Direction	Degrees Magnetic	N/A	Calculated	TAS+HDG+GS+TRK
HDG	Heading	Degrees Magnetic	Syn/Step/Man.	Compass System or Pilot Input	Slaved Compass System
DRE	Drift	Degrees	N/A	Calculated	TAS+HDG+GS+TRK
WDY/WS	Wind Direction/Speed	Degrees Magnetic/Knots	N/A	Calculated	TAS+HDG+GS+TRK
HWD	Head/Tail Wind	Knots	N/A	Calculated	TAS+HDG+GS+TRK
HWD	Head/Tail Wind	Miles per Hour	N/A	Calculated	TAS+HDG+GS+TRK
EF	Engine Efficiency	Miles per Gallon	N/A	Calculated	FF+GS
TAT	Total Air Temperature	Degrees Celsius	N/A	Measurement of OAT Sensor	OAT Sensor
TAT	Total Air Temperature	Degrees Fahrenheit	N/A	Measurement of OAT Sensor	OAT Sensor
SAT	Static Air Temperature	Degrees Celsius	N/A	Calculated	TAT+MACH
SAT	Static Air Temperature	Degrees Fahrenheit	N/A	Calculated	TAT+MACH
ISA	ISA Temperature	Degrees Celsius	N/A	Calculated	AL+SAT
OV	Sensor Temperature	Degrees Celsius	N/A	Internal Measurement	
TR	Total Fuel Remaining	Configurable Units	LIT/GAL/LB	Pilot Input+Calculation+FF	FF Sensor+K Factor
TU	Total Fuel Used	Configurable Units	LIT/GAL/LB	Calculated from FF	FF Sensor+K Factor
TF	Total Fuel Flow	Configurable Units per Hour	LIT/GAL/LB	Measurement of FF	FF Sensor+K Factor
FL	Left Engine Fuel Flow	Configurable Units per Hour	LIT/GAL/LB	Measurement of FF	FF Sensor+K Factor
FR	Right Engine Fuel Flow	Configurable Units per Hour	LIT/GAL/LB	Measurement of FF	FF Sensor+K Factor
UL	Left Engine Fuel Used	Configurable Units	LIT/GAL/LB	Calculated from FF	FF Sensor+K Factor
UR	Right Engine Fuel Used	Configurable Units	LIT/GAL/LB	Calculated from FF	FF Sensor+K Factor
FU/FR	Fuel Flow Left and Right	Configurable Units per Hour	LIT/GAL/LB	FF Sensor(s)	FF Sensor+K Factor
BT	Battery Voltage	Volts	N/A	Internal Measurement	Supply Bus Voltage
AA	Altitude Alert	Feet	AWB	Pilot Input+AL+AWB	Expansion Module
AD	Altitude Difference	Feet	N/A	Calculated from AL+AA	AL+AA
OVS	Overspeed Warning	Knots	Factory Preset	Factory Preset+TAS	Expansion Module
N/A	K Factor(s), KL, KR	Pulses per Gallon	default=30,000	Set during installation	Internal Memory
N/A	Maximum Fuel/Onboard	Gallons	Dealer Set	Set during installation	Internal Memory

TAS 1000 MODIFICATION DRAWING LIST #1000-403 Rev H

Drawing Number	Rev.	Drawing Title
1000-001	D	TAS 1000 AIR DATA COMPUTER
1000-002	C	WINDICATOR
1000-301	J	TAS 1000 INSTALLATION SCHEMATIC Pages 1-4
1000-302	-	OAT MOUNTING DIAGRAM
1000-303	D	FUEL FLOW / TF500 SENSOR MOUNTING DIAGRAM 1-3
1000-304	-	TAS 1000 PANEL CUT-OUT
1000-305	B	WINDICATOR MOUNTING DIAGRAM Pages 1-3
1000-816	A	TF500 SENSOR INSTALLATION SCHEMATIC

The TAS installation must be performed in accordance with Advisory Circular 43.13-1 B Acceptable Methods, Techniques, and Practices - Aircraft Inspection and Repair.

REVISIONS		
REV	DESCRIPTION	DATE
A	Added Specifications and Limitations.	FEB 17/00
B	Added test and calibration label. See ECN #TAS003.	OCT 10/00
C	Added modification level to drawing	APR 8/03
D	Added modification level to drawing	AUG 9/05

SIDE VIEW

FRONT VIEW

SPECIFICATIONS AND LIMITATIONS

SYSTEM COMPONENTS:

- 1000-001 PANEL MOUNT COMPUTER
- 1000-002 INDICATOR (Optional)
- 1000-003 EXPANSION MODULE (ALTITUDE ENCODER, OVERSPEED WARNING) (Optional)
- 1000-004 FUEL FLOW SENSOR (1 OR 2 Optional)
- 1000-006 TF500 FLOW SENSOR (Optional)

Power: 10 TO 30 VDC, 30 W Maximum, 5 W Continuous.
Weight: 1.0 lb.
Fuel Flow Rate: 1.0 to 400 GPH.
Maximum Fuel Quantity: 2000 U.S.G. or 9999 lb.
Altitude: -1000 to 55000 ft.
Maximum Airspeed: 500 knots.
Operating Temperature: -30° C to +60° C.
Accuracy: better than 1.5 %.
Serial Interface: RS-232.
Environmental: DO-160D Env. Cat. F1BABSXXXXXXXXZBBBBBXXXXXX.
Software: DO-178B Level C.
TSO: C106, C44b, C101, C88a.

BOTTOM VIEW

TEST AND CALIBRATION LABEL

Insight

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TRUE AIR DATA COMPUTER SYSTEM

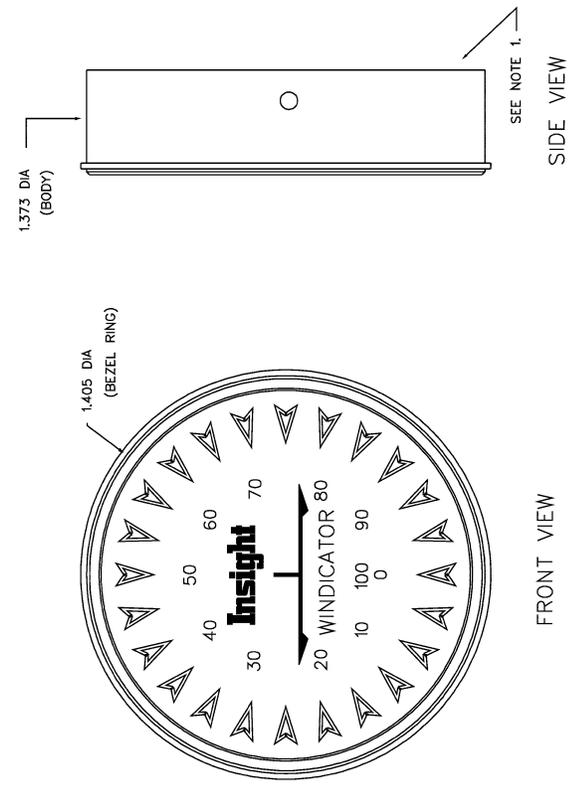
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SIZE	DRAWING NO.	REV
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SCALE	DATE	SHEET
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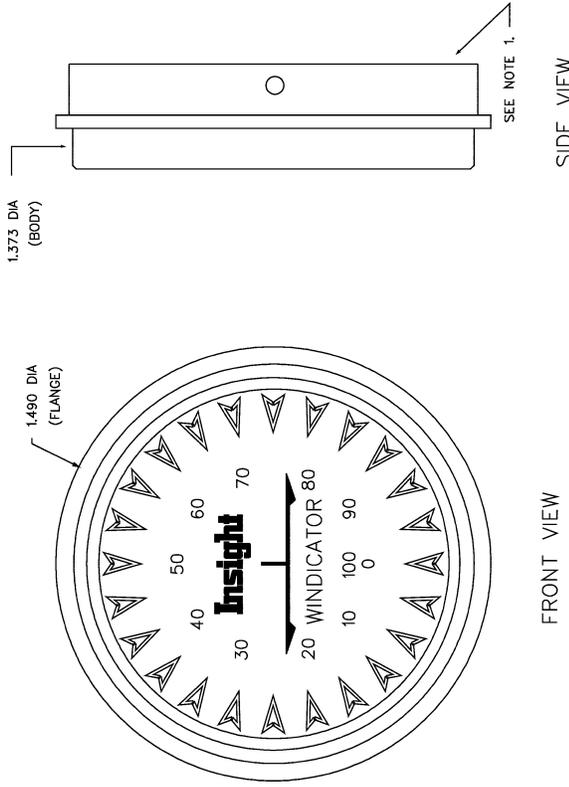
ALL DIMENSIONS IN INCHES, TOLERANCE +/- 0.050.

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REVISIONS			
REV	DESCRIPTION	DATE	APPROVED
A	CHANGED BEZEL DIM. FROM 1.366 TO 1.373	Sept 1/99	JWT
B	ADDED TYPE B	June 22/00	JWT
C	REMOVED PRE-TERMINATED CABLE AT WINDICATOR	June 18/02	RR



1000-002-A
(FRONT MOUNT)



1000-002-B
(REAR MOUNT)

NOTES:

1. P/N 1000-503 WINDICATOR NAMEPLATE.
2. All dimensions in inches, tolerance +/- 0.010 U.O.S.

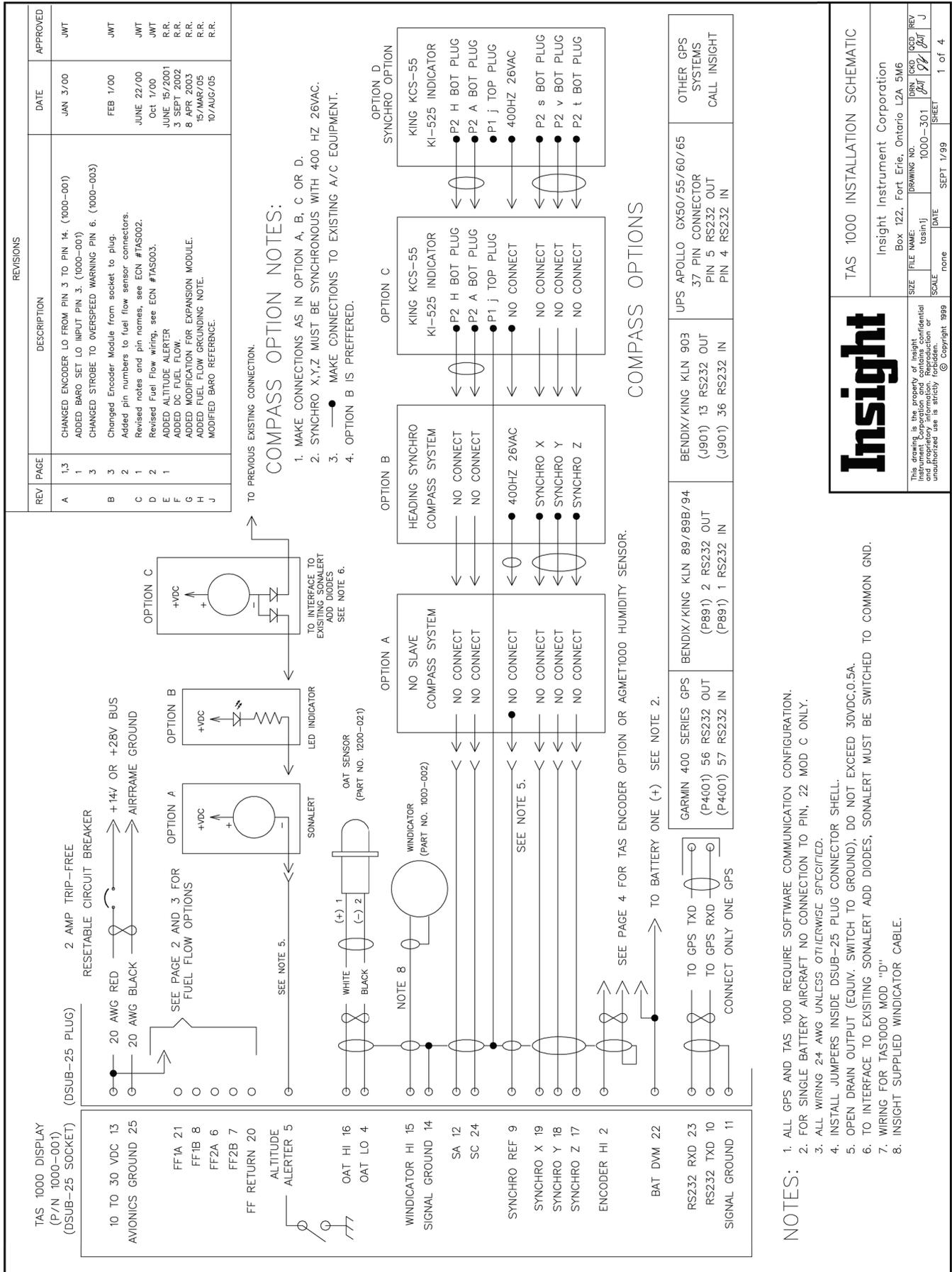
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SIZE	FILE NAME: WIn 1000c	DWG NO. 1000-002	DWN I/C/D	REV
SCALE	NONE	DATE	June 1/99	SHEET 1 of 1

WINDICATOR

Insight Instrument Corporation
Box 122, Fort Erie, Ontario L2A 5M6



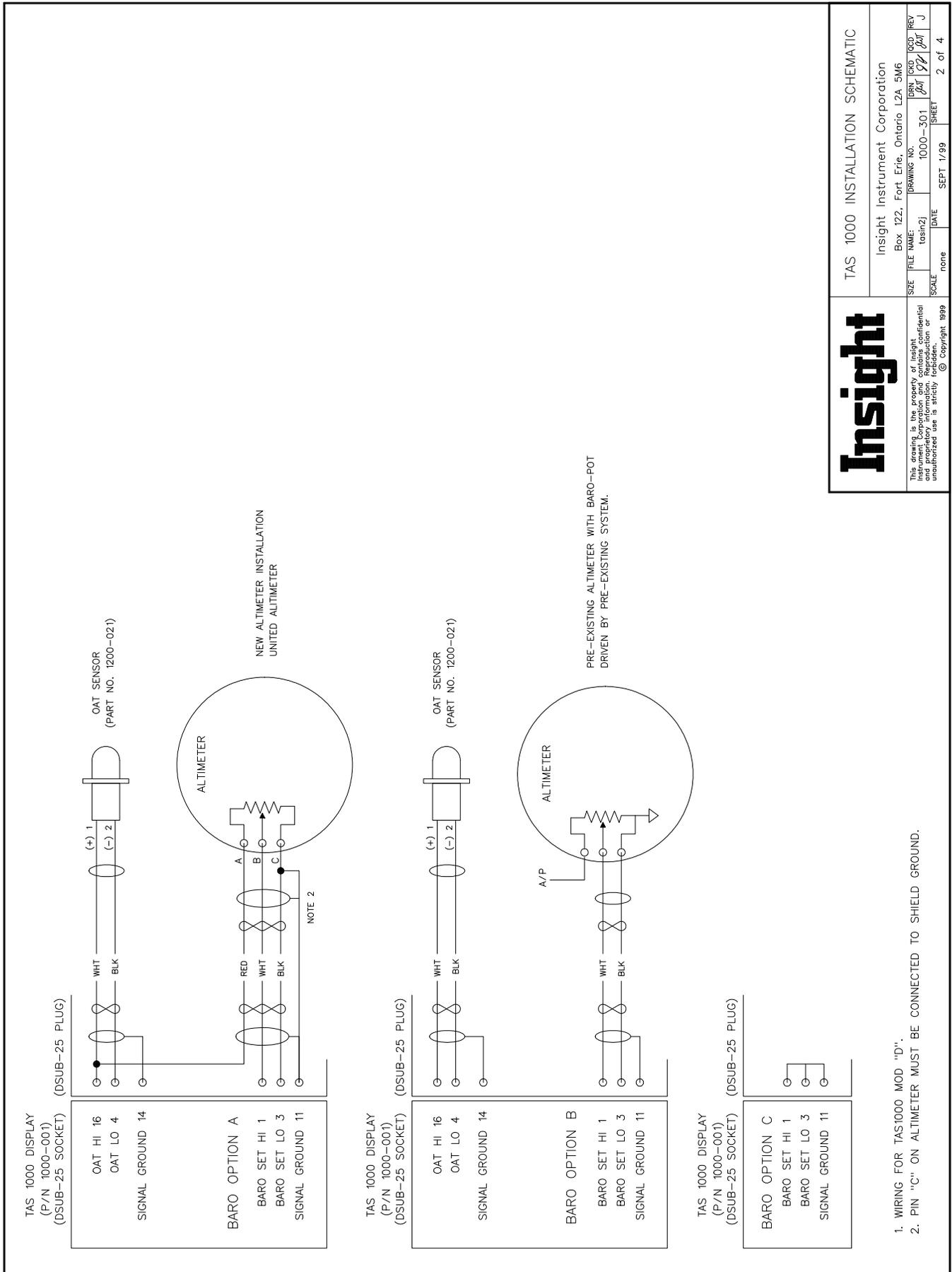
- NOTES:**
1. ALL GPS AND TAS 1000 REQUIRE SOFTWARE COMMUNICATION CONFIGURATION.
 2. FOR SINGLE BATTERY AIRCRAFT NO CONNECTION TO PIN, 22 MOD C ONLY.
 3. ALL WIRING 24 AWG UNLESS OTHERWISE SPECIFIED.
 4. INSTALL JUMPERS INSIDE DSUB-25 PLUG CONNECTOR SHELL.
 5. OPEN DRAIN OUTPUT (EQUIV. SWITCH TO GROUND), DO NOT EXCEED 30VDC,0.5A.
 6. TO INTERFACE TO EXISTING SONALERT ADD DIODES, SONALERT MUST BE SWITCHED TO COMMON GND.
 7. WIRING FOR TAS1000 MOD "D".
 8. INSIGHT SUPPLIED WINDICATOR CABLE.

Insight

TAS 1000 INSTALLATION SCHEMATIC

Insight Instrument Corporation
Box 122, Fort Erie, Ontario L2A 5M6

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	FILE NAME:	1000-301			
	SCALE	DATE	SEPT 1/99	SHEET	1 of 4
	none				



Insight

TAS 1000 INSTALLATION SCHEMATIC

Insight Instrument Corporation
Box 122, Fort Erie, Ontario L2A 5M6

FILE NAME: tasin2j
DRAWING NO. 1000-301
SCALE: none
DATE: SEPT 1/99
SHEET: 2 of 4

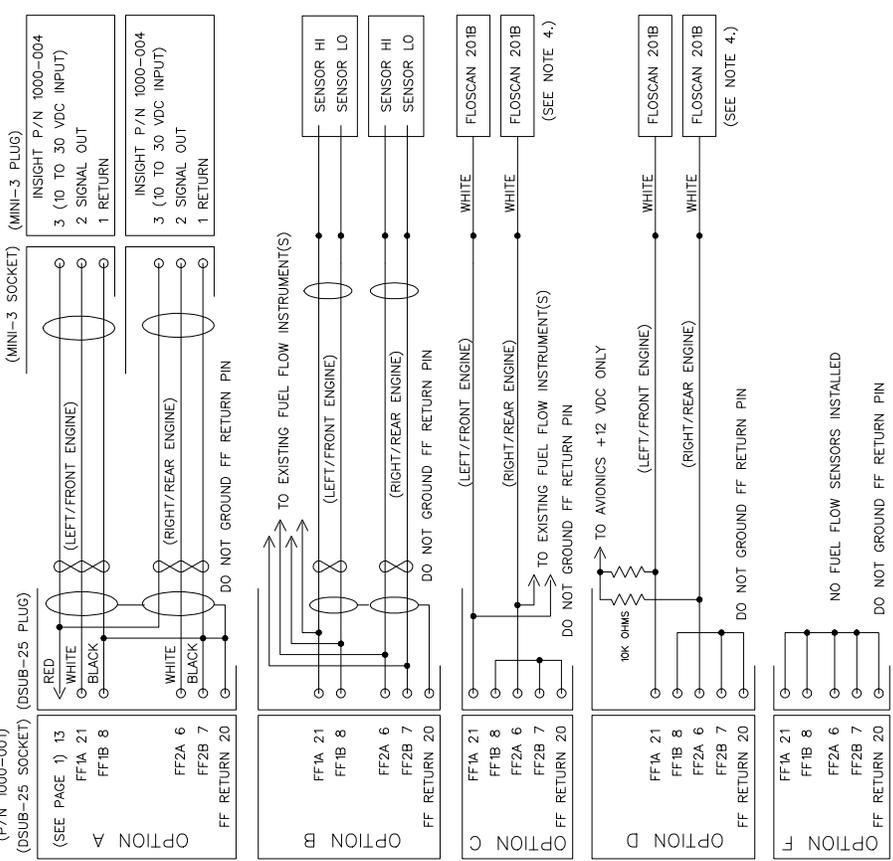
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SEE PAGE ONE FOR REVISION LOG

FUEL FLOW SENSOR CONNECTIONS

(TWIN ENGINE INSTALLATION SHOWN)
(SEE NOTE 1 FOR SINGLE ENGINE INSTALLATION)

TAS 1000 DISPLAY
(P/N 1000-001)
(DSUB-25 SOCKET)



INSIGHT FUEL FLOW SENSORS INSTALLED (SEE NOTE 6.)

SINEMAVE AC FUEL FLOW EXISTING SENSORS CONNECTED TO EXISTING FUEL FLOW INSTRUMENT

FLOSCAN 201 FUEL FLOW EXISTING SENSORS CONNECTED TO EXISTING FUEL FLOW INSTRUMENT (10K PULL-UPS NOT SHOWN)

FLOSCAN 201 FUEL FLOW EXISTING SENSORS CONNECTED ONLY TO INSIGHT TAS SYSTEM

NOTES:

1. SINGLE ENGINE AIRCRAFT USE FF1 INPUTS, TERMINATE FF2A AND FF2B AS IN OPTION F.
2. ALL WIRING 24 AWG U.O.S., INSTALL JUMPERS INSIDE DSUB-25 CONNECTOR SHELL.
3. ALL FUEL FLOW INPUTS (PINS 6, 7, 8, 21) MUST BE TERMINATED.
4. TERMINATE FLOSCAN 201B RED WIRE TO FUSED 12VDC, BLACK WIRE TO AIRFRAME GROUND. (ALL FLOSCAN 201B SENSORS REQUIRE 10K PULL-UP FROM WHITE WIRE TO 12VDC)
5. WIRING FOR TAS1000 MOD D.
6. FOR INSIGHT FUEL FLOW SENSORS, THE BRACKET MUST BE CONNECTED TO AIRFRAME GROUND.

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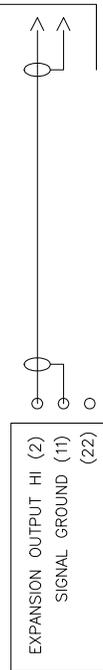
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Insight Instrument Corporation Box 122, Fort Erie, Ontario L2A 5M6	
FILE NAME: tesin3i	DRAWING NO: 1000-301
SCALE: none	DATE: SEPT 1/99
SIZE: A	REV: J
REV: J	SHEET: 3 of 4

TAS EXPANSION MODULE CONNECTIONS

SEE PAGE ONE FOR REVISION LOG

TAS EXPANSION
(P/N 1000-003)
(DSUB-15 PLUG) (H1)

TAS 1000 DISPLAY (PART NO. 1000-001)
(DSUB-25 SOCKET)



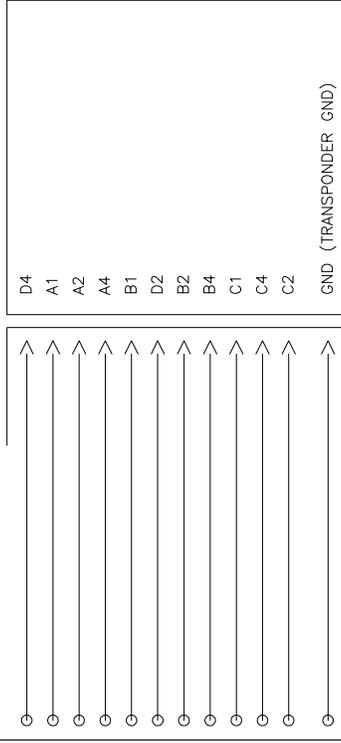
OPTION 1
NOTE 8

- 8 D4
- 7 A1
- 6 A2
- 5 A4
- 4 B1
- 3 D2
- 15 B2
- 14 B4
- 13 C1
- 12 C4
- 11 C2
- 2 PGND

1 OVERSPEED WARNING

(DSUB-15 SOCKET CONT)

TRANSPONDER (OR OTHER GRAY-CODE RECEIVER)
(SEE NOTE 7) (SEE INSTALLATION PROCEDURES)



+12/28 VDC
(FUSED OR C.B. PROTECTED)

(SEE NOTE 7)

OVERSPEED WARNING DEVICE

- DEVICE HI
- DEVICE LO

SEE NOTE 6.

NOTES:

1. NO CONNECTION TO TAS 1000 PIN 2 IF TAS EXPANSION IS NOT INSTALLED.
2. NO CONNECTION TO TAS EXPANSION PIN 2.
3. ALL WIRING 24 AWG UNLESS OTHERWISE SPECIFIED.
4. INSTALL JUMPERS INSIDE DSUB15 CONNECTOR SHELL.
5. ISOLATION DIODES ARE RECOMMENDED FOR ALL GRAY-CODE RECEIVING DEVICES.
SOME BUT NOT ALL GRAY-CODE RECEIVERS CONTAIN INTERNAL ISOLATION DIODES.
FOR GRAY-CODE RECEIVERS WITHOUT INTERNAL ISOLATION DIODES, INSERT A 1N4007 OR EQUIVALENT IN SERIES WITH EACH LINE (ANODE TO RECEIVER).
6. CONNECT TO OVERSPEED WARNING OUTPUT ONLY IF PART NO. 1000-003 MATCHES AIRCRAFT TYPE AND CONFIGURATION.
OVERSPEED WARNING OUTPUT IS NORMALLY OPEN (HI-Z) AND SWITCHES TO GROUND DURING OVERSPEED CONDITION.
OVERSPEED WARNING OUTPUT RATED 28VDC 0.25 AMPS MAXIMUM. (SEE INSTALLATION MANUAL)
7. INSTALLER PROVIDED CONNECTORS.
8. EXPANSION MODULE 1000-004 MOD.B

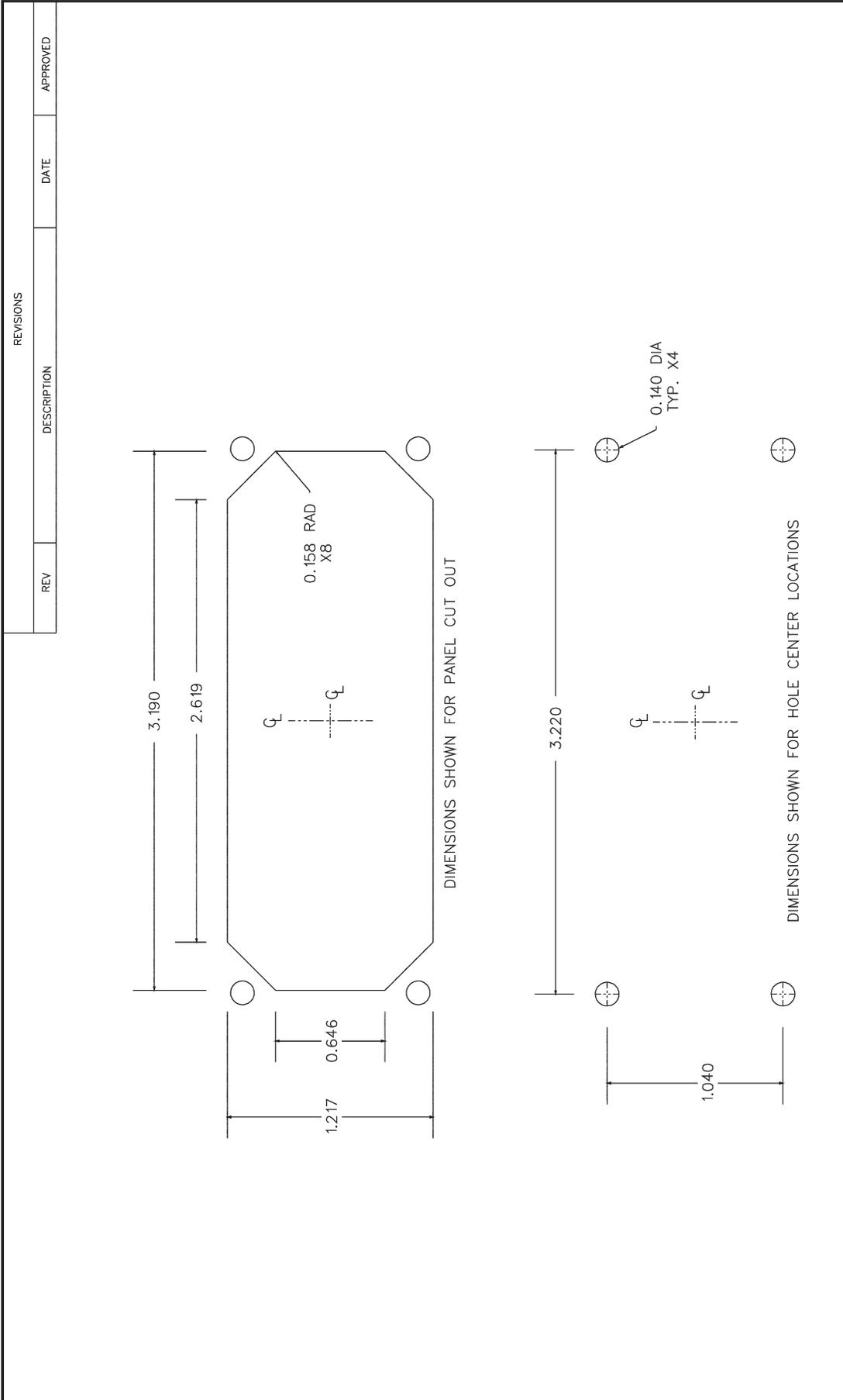


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TAS 1000 INSTALLATION SCHEMATIC

Insight Instrument Corporation
Box 122, Fort Erie, Ontario L2A 5M6

SIZE	FILE NAME	DRAWING NO.	REV	ESD	ESD	REV
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SCALE	DATE	SHEET	4 of 4			
none	SEPT 1/99					



Insight

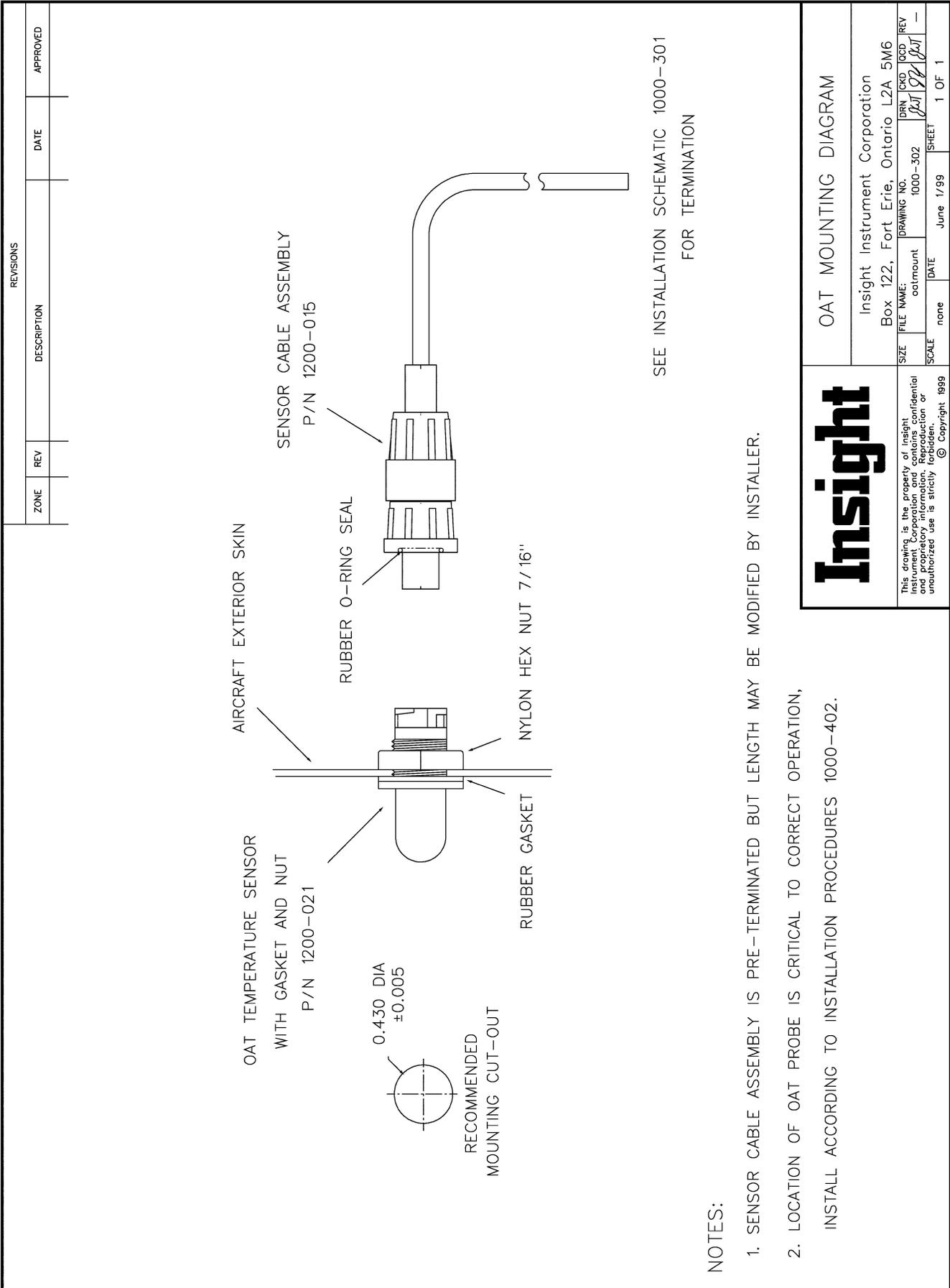
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TAS 1000 PANEL CUT-OUT DIAGRAM

Insight Instrument Corporation
Box 122, Fort Erie, Ontario L2A 5M6

SIZE	FILE NAME:	DWG NO.	DRN	CHK	REV
	BW2	1000-304			
SCALE	NONE	DATE	OCT 13/99	SHEET	1 of 1

- NOTES:
- 1: CUTOUT DIMENSIONS SHOWN FOR .010 CLEARANCE FIT.
 - 2: ALL DIMENSIONS SHOWN ARE SYMMETRICAL AROUND THE CENTER POINT.
 - 3: NOT TO SCALE, DO NOT USE AS A TEMPLATE.



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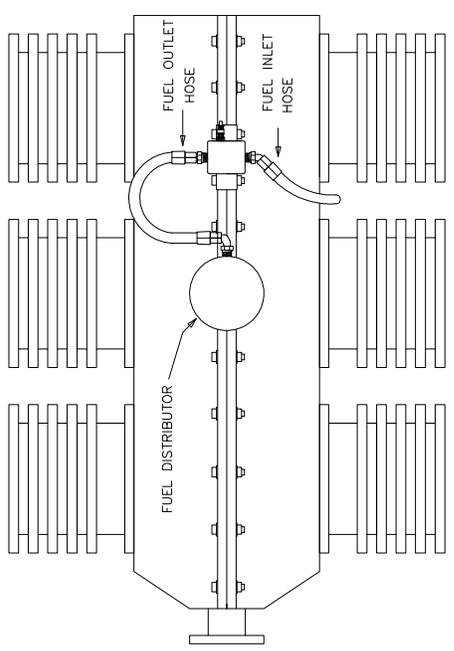
OAT MOUNTING DIAGRAM

Insight Instrument Corporation

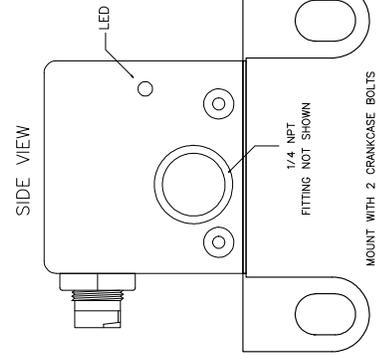
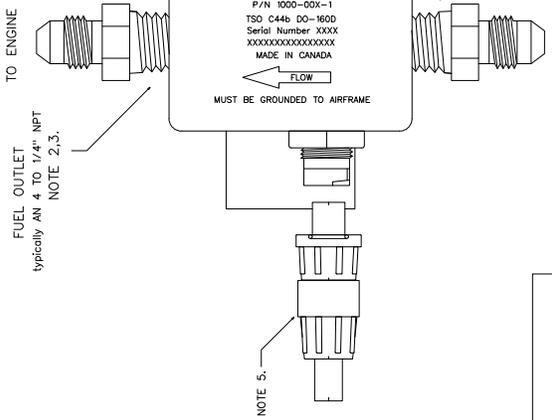
Box 122, Fort Erie, Ontario L2A 5M6

SIZE	DRAWING NO.	DIRN	ICD	REV
	1000-302	001	001	—
SCALE	none	DATE	June 1/99	SHEET 1 OF 1

REVISIONS			
REV	DESCRIPTION	DATE	APPROVED
A	Added housing and bracket.	June 22/00	JWT
B	CHANGED TITLE	June 15/01	R.R.
C	ADDED 2 ADDITIONAL MOUNTING OPTIONS	MAY 26/03	R.R.
D	ADDED VIBRATION AND K-FACTOR OPTIONS	DEC 10/10	JY



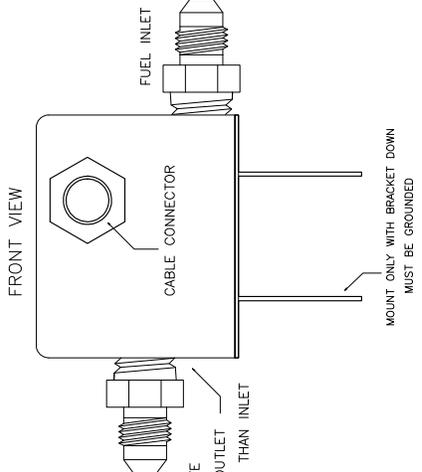
TYPICAL INSTALLATION



PART NUMBERS		
INJECTOR	CARBURETOR	
FLOW SENSOR	1000-004-1(-Kmmnn)(-EPR)	1000-004-4(-Kmmnn)(-EPR)
TRUE FLOW SENSOR	1000-006-1X SEE NOTE 12.	1000-006-4X SEE NOTE 12.
FLOW/VIBRATION SENSOR	1000-004A-1	1000-004A-4
VIBRATION SENSOR	1000-004V(-Xn) SEE NOTE 15.	1000-004V(-Xn) SEE NOTE 15.

NOTES:

1. Install INSIGHT FLOW SENSOR, one per engine.
2. Use ONLY approved 1/4 NPT fittings for transducer inlet and outlet, installer supplied.
3. Use ONLY approved fuel line assemblies.
4. Use approved thread sealant. Torque pipe threads to a max of 15 ft. lbs. (180 in. lbs.)
5. INSIGHT supplied electrical harness; 1/3 TURN, 3-pin OR clip-on 7-pin for FLOW/VIBRATION SENSOR.
6. A straight length of fuel line upstream from the sensor inlet will improve accuracy by reducing turbulence in the rotor.
7. CONFIRM TRANSDUCER CASE IS GROUNDED TO AIRFRAME GROUND BEFORE POWERING UP.
8. Mount transducer with nomenclature facing up.
9. Confirm all wiring PRIOR to powering up.
10. SEE INSTALLATION SCHEMATIC 1000-301 FOR FLOW SENSOR WIRING DETAILS.
11. SEE INSTALLATION SCHEMATIC 1000-816 FOR TRUE FLOW SENSOR WIRING DETAILS.
12. SPECIFY GPS MANUFACTURE AND MODEL WHEN ORDERING TRUE FLOW SENSOR, BENDIX KING B-FORMAT, APOLLO AND GARMIN Z-FORMAT.
13. FLOW SENSOR 30,000 PULSES PER GALLON UNLESS INDICATED BY -Kmmnn.
14. FLOW SENSOR DATA SIGNAL 5V UNLESS -EPR, EXTERNAL PULL-UP RESISTOR REQUIRED. SEE WIRING DIAGRAM.
15. Install vibration sensor, one per engine, consult insight for alternate application.



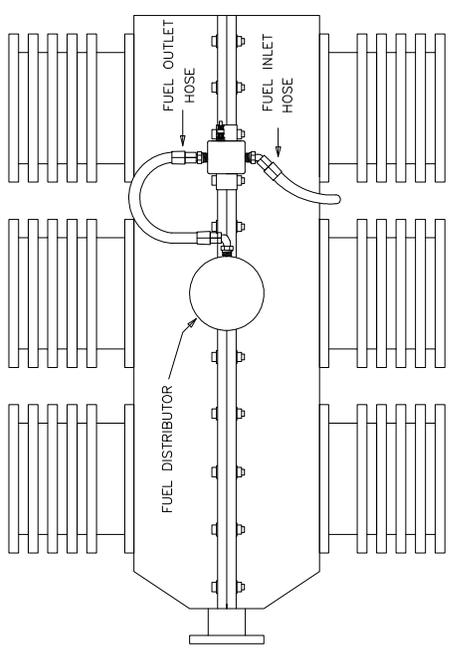
PIN	FUNCTION
1	AVIONICS GROUND
2	DATA OUT (TRANSMIT)
3	10 TO 30 VDC INPUT

PIN	FUNCTION
1	10 TO 30 VDC INPUT
2	DATA OUT (TRANSMIT)
3	AVIONICS GROUND
4	RESERVED
5	C2 CLOCK
6	C2 DATA
7	DATA IN (RECEIVE)

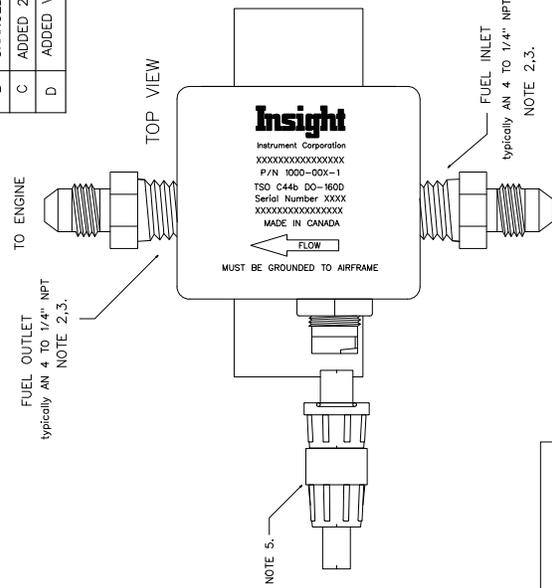
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FLOW SENSOR AND TRUE FLOW SENSOR MOUNTING DIAGRAM (BRACKET TYPE 1)			
Insight Instrument Corporation Box 122, Fort Erie, Ontario L2A 5M6			
SIZE	FILE NAME:	DRAWING NO.:	DATE:
SCALE	none	1000-303	June 1/99
			SHEET 1 OF 3

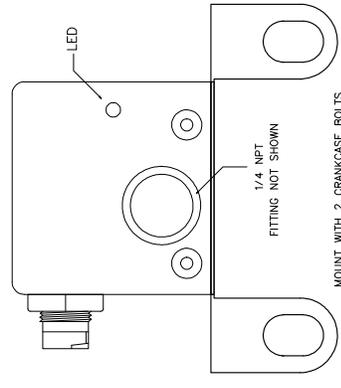
REVISIONS			
REV	DESCRIPTION	DATE	APPROVED
A	Added housing and bracket.	June 22/00	JWT
B	CHANGED TITLE	June 15/01	R.R.
C	ADDED 2 ADDITIONAL MOUNTING OPTIONS	MAY 26/03	R.R.
D	ADDED VIBRATION AND K-FACTOR OPTIONS	DEC 10/10	JY



FUEL OUTLET
Typically AN 4 TO 1/4" NPT
NOTE 2.3.

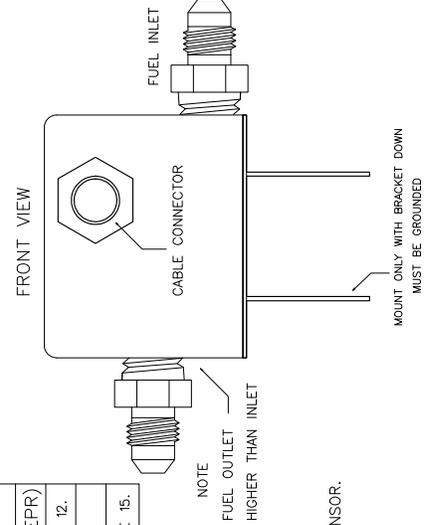


SIDE VIEW



TYPICAL INSTALLATION

PART NUMBERS	
INJECTOR	CARBURETOR
FLOW SENSOR	1000-004-1(-Kmmnn)(-EPR)
TRUE FLOW SENSOR	1000-006-1X SEE NOTE 12.
FLOW/VIBRATION SENSOR	1000-004A-1
VIBRATION SENSOR	1000-004V(-Xn) SEE NOTE 15.



FRONT VIEW

PIN	FUNCTION
1	AVIONICS GROUND
2	DATA OUT (TRANSMIT)
3	10 TO 30 VDC INPUT

3-PIN CONNECTOR PINOUT

PIN	FUNCTION
1	10 TO 30 VDC INPUT
2	DATA OUT (TRANSMIT)
3	AVIONICS GROUND
4	RESERVED
5	C2 CLOCK
6	C2 DATA
7	DATA IN (RECEIVE)

7-PIN CONNECTOR PINOUT

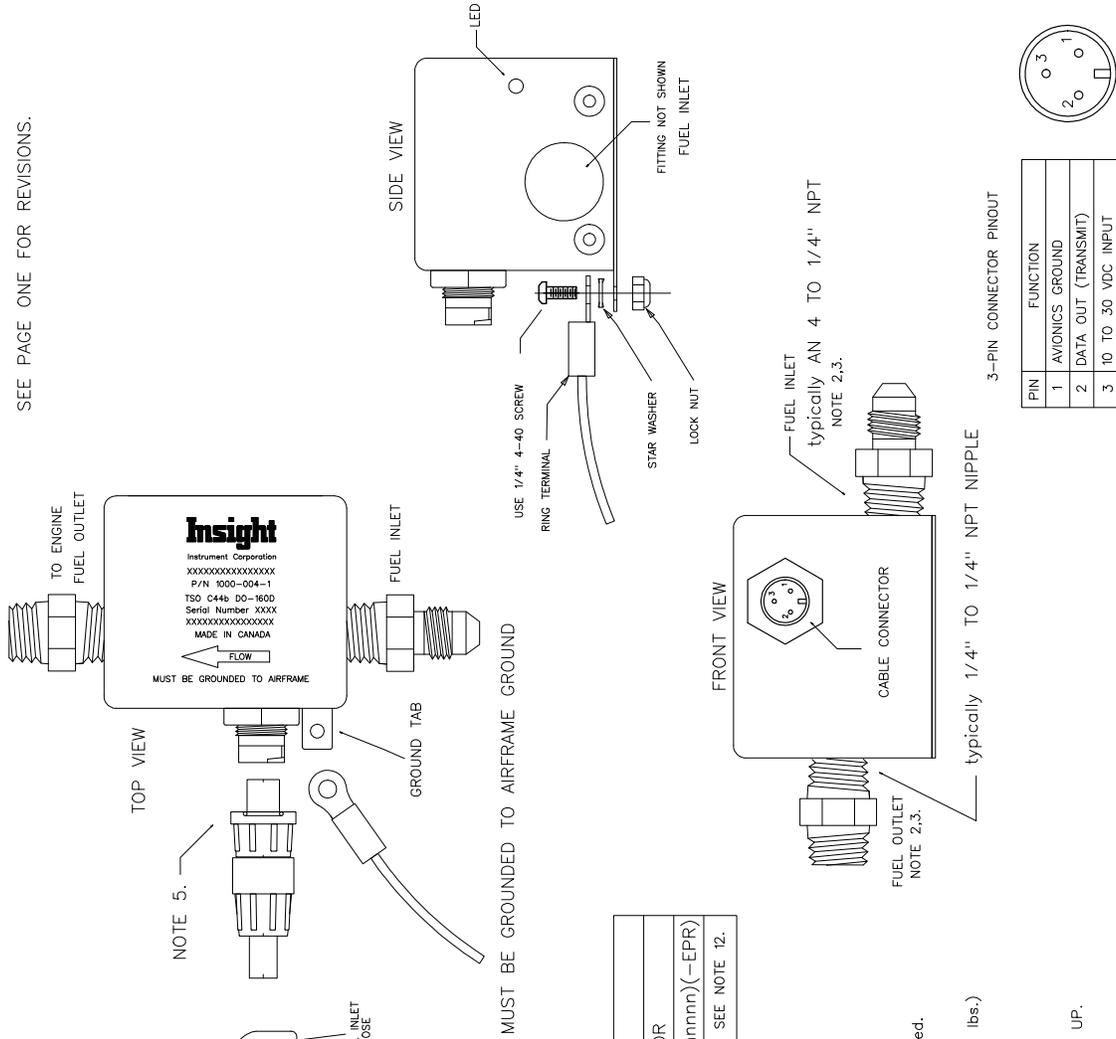
NOTES:

1. Install INSIGHT FLOW SENSOR, one per engine.
2. Use ONLY approved 1/4 NPT fittings for transducer inlet and outlet, installer supplied.
3. Use ONLY approved fuel line assemblies.
4. Use approved thread sealant. Torque pipe threads to a max of 15 ft. lbs. (180 in. lbs.)
5. INSIGHT supplied electrical harness; 1/3 TURN, 3-pin OR clip-on 7-pin for FLOW/VIBRATION SENSOR.
6. A straight length of fuel line upstream from the sensor inlet will improve accuracy by reducing turbulence in the rotor.
7. CONFIRM TRANSDUCER CASE IS GROUNDED TO AIRFRAME GROUND BEFORE POWERING UP.
8. Mount transducer with nomenclature facing up.
9. Confirm all wiring PRIOR to powering up.
10. SEE INSTALLATION SCHEMATIC 1000-301 FOR FLOW SENSOR WIRING DETAILS.
11. SEE INSTALLATION SCHEMATIC 1000-816 FOR TRUE FLOW SENSOR WIRING DETAILS.
12. SPECIFY GPS MANUFACTURE AND MODEL WHEN ORDERING TRUE FLOW SENSOR, BENDIX KING B-FORMAT, APOLLO AND GARMIN Z-FORMAT.
13. FLOW SENSOR 30,000 PULSES PER GALLON UNLESS INDICATED BY -Kmmnn.
14. FLOW SENSOR DATA SIGNAL 5V UNLESS -EPR, EXTERNAL PULL-UP RESISTOR REQUIRED. SEE WIRING DIAGRAM.
15. Install vibration sensor, one per engine, consult insight for alternate application.

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FLOW SENSOR AND TRUE FLOW SENSOR MOUNTING DIAGRAM (BRACKET TYPE 1)	
Insight Instrument Corporation Box 122, Fort Erie, Ontario L2A 5M6	
SIZE	FILE NAME: insight
DRAWING NO:	1000-402
DATE	June 1/99
SCALE	none
SHEET	1 OF 3

SEE PAGE ONE FOR REVISIONS.

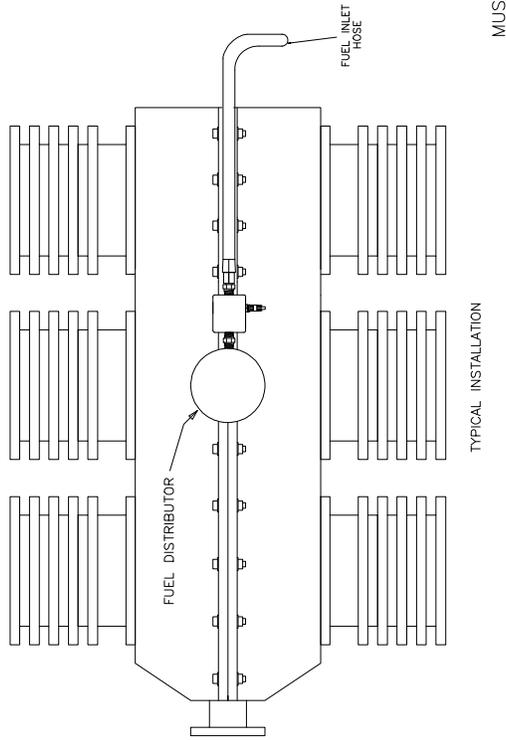


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FILE NAME: 1000-303
 DRAWING NO.: 1000-303
 DATE: June 1/79
 SHEET: 3 OF 3

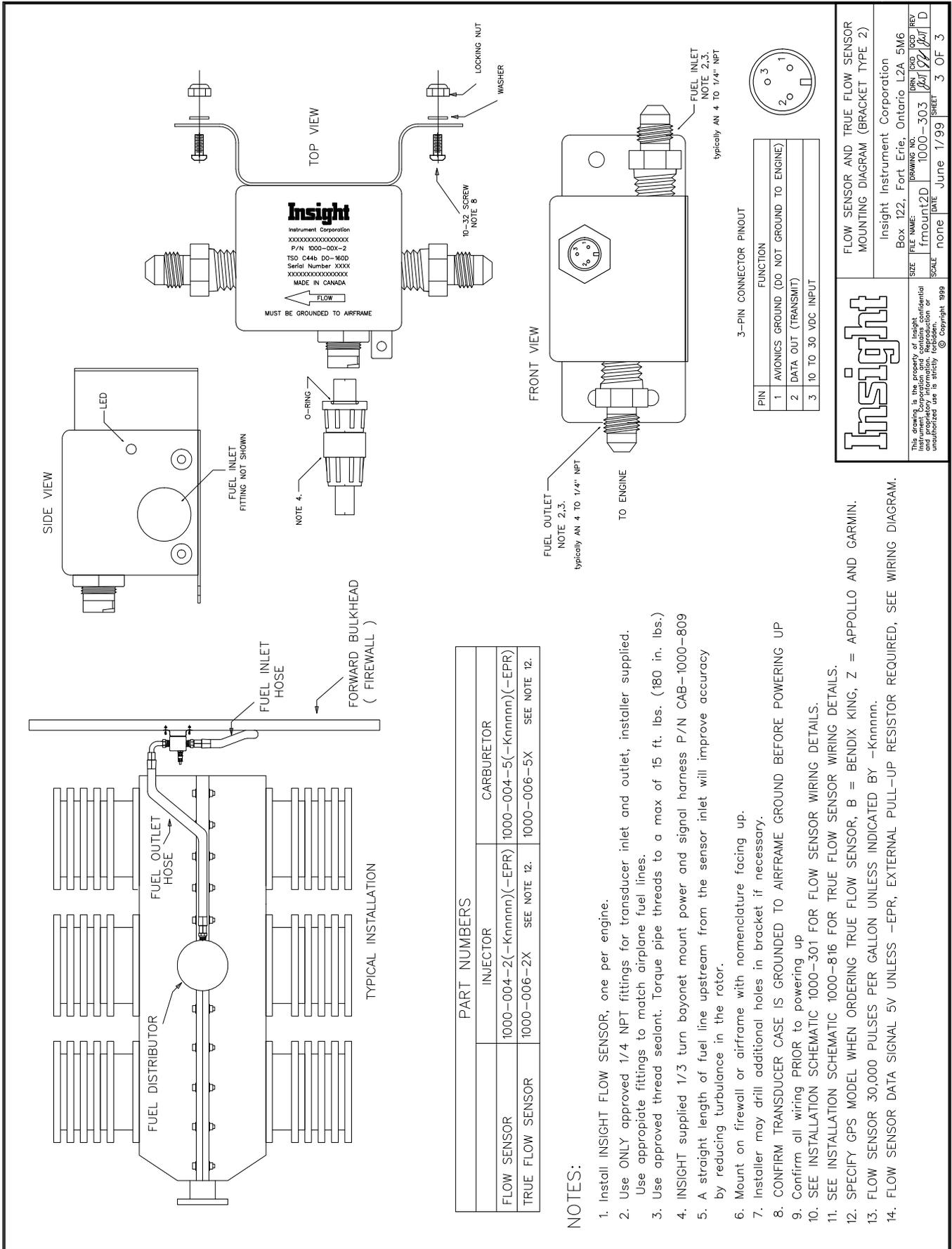
FLOW SENSOR AND TRUE FLOW SENSOR MOUNTING DIAGRAM (BRACKET TYPE 3)
 Insight Instrument Corporation
 Box 122, Fort Erie, Ontario L2A 5M6



PART NUMBERS	
FLOW SENSOR	CARBURETOR
1000-004-3(-Knnnn)(-EPR) SEE NOTE 12.	1000-004-6(-Knnnn)(-EPR)
1000-006-3X SEE NOTE 12.	1000-006-6X SEE NOTE 12.

NOTES:

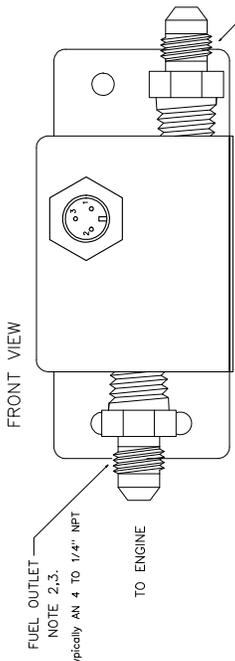
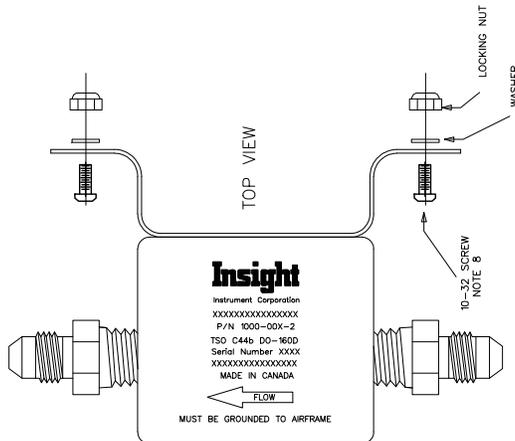
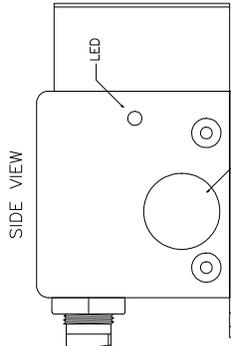
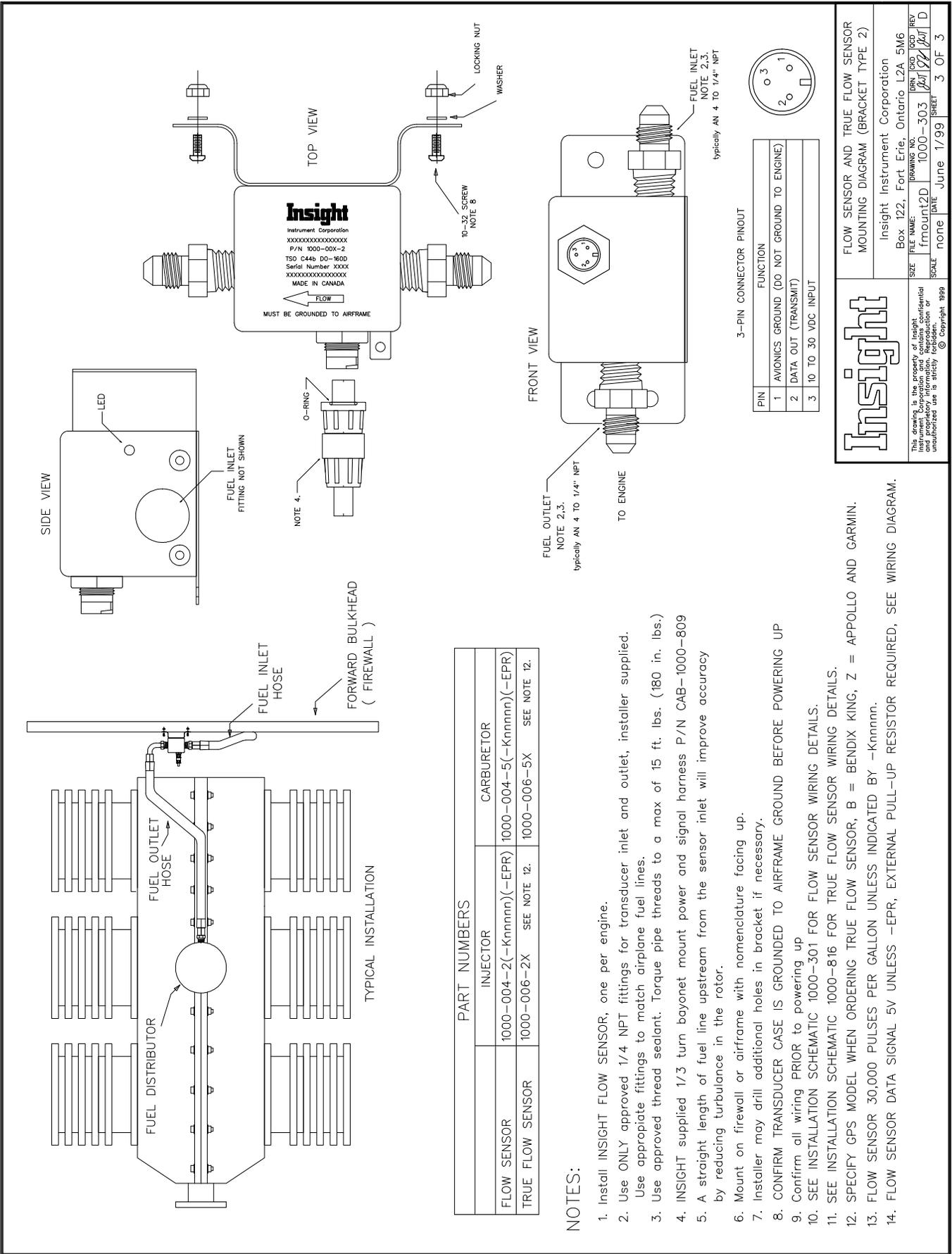
1. Install INSIGHT FLOW SENSOR, one per engine.
2. Use ONLY approved 1/4 NPT fittings for transducer inlet and outlet, installer supplied.
3. Use ONLY approved fuel line assemblies.
4. Use approved thread sealant. Torque pipe threads to a max of 15 ft. lbs. (180 in. lbs.)
5. INSIGHT supplied 1/3 turn bayonet electrical harness P/N CAB-1000-809.
6. A straight length of fuel line upstream from the sensor inlet will improve accuracy by reducing turbulence in the rotor.
7. CONFIRM TRANSDUCER CASE IS GROUNDED TO AIRFRAME GROUND BEFORE POWERING UP.
8. Mount transducer with nomenclature facing up.
9. Confirm all wiring PRIOR to powering up.
10. SEE INSTALLATION SCHEMATIC 1000-301 FOR FLOW SENSOR WIRING DETAILS.
11. SEE INSTALLATION SCHEMATIC 1000-816 FOR TRUE FLOW SENSOR WIRING DETAILS.
12. SPECIFY GPS MANUFACTURE AND MODEL WHEN ORDERING TRUE FLOW SENSOR, BENDIX KING B-FORMAT, APOLLO AND GARMIN Z-FORMAT.
13. FLOW SENSOR 30,000 PULSES PER GALLON UNLESS INDICATED BY -Knnnn.
14. FLOW SENSOR DATA SIGNAL 5V UNLESS -EPR, EXTERNAL PULL-UP RESISTOR REQUIRED, SEE WIRING DIAGRAM.
15. Transducer self supporting on fuel fittings.



PART NUMBERS	
FLOW SENSOR	TRUE FLOW SENSOR
1000-004-2(-Knnmm)(-EPR)	1000-004-5(-Knnmm)(-EPR)
1000-006-2X SEE NOTE 12.	1000-006-5X SEE NOTE 12.

NOTES:

1. Install INSIGHT FLOW SENSOR, one per engine.
2. Use ONLY approved 1/4 NPT fittings for transducer inlet and outlet, installer supplied.
3. Use appropriate fittings to match airplane fuel lines.
4. Use approved thread sealant. Torque pipe threads to a max of 15 ft. lbs. (180 in. lbs.)
5. INSIGHT supplied 1/3 turn bayonet mount power and signal harness P/N CAB-1000-809 by reducing turbulence in the rotor.
6. Mount on firewall or airframe with nomenclature facing up.
7. Installer may drill additional holes in bracket if necessary.
8. CONFIRM TRANSDUCER CASE IS GROUNDED TO AIRFRAME GROUND BEFORE POWERING UP
9. Confirm all wiring PRIOR to powering up
10. SEE INSTALLATION SCHEMATIC 1000-301 FOR TRUE FLOW SENSOR WIRING DETAILS.
11. SEE INSTALLATION SCHEMATIC 1000-816 FOR TRUE FLOW SENSOR WIRING DETAILS.
12. SPECIFY GPS MODEL WHEN ORDERING TRUE FLOW SENSOR, B = BENDIX KING, Z = APOLLO AND GARMIN.
13. FLOW SENSOR 30,000 PULSES PER GALLON UNLESS INDICATED BY -Knnmm.
14. FLOW SENSOR DATA SIGNAL 5V UNLESS -EPR, EXTERNAL PULL-UP RESISTOR REQUIRED, SEE WIRING DIAGRAM.



PIN	FUNCTION
1	AVIONICS GROUND (DO NOT GROUND TO ENGINE)
2	DATA OUT (TRANSMIT)
3	10 TO 30 VDC INPUT

INSIGHT

Flow Sensor and True Flow Sensor Mounting Diagram (Bracket Type 2)

Insight Instrument Corporation
Box 122, Fort Erie, Ontario L2A 5M6

SIZE: DRAWING NO. 1000-303
FILE NAME: fmount2D
SCALE: none DATE: June 1/99 SHEET: 3 OF 3

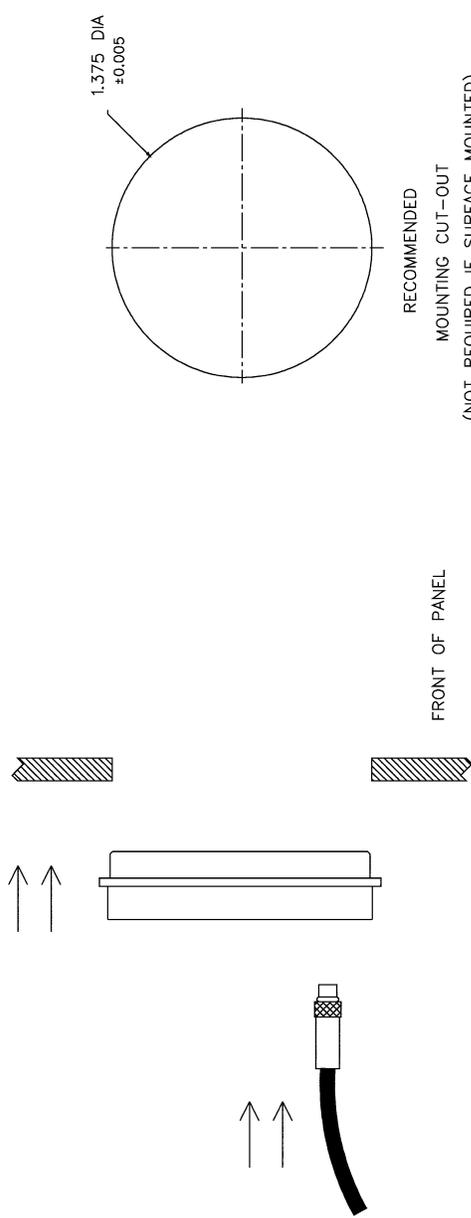
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PART NUMBERS	
FLOW SENSOR	CARBURETOR
1000-004-2(-Knnmm)(-EPR) SEE NOTE 12.	1000-004-5(-Knnmm)(-EPR)
1000-006-2X	1000-006-5X

NOTES:

1. Install INSIGHT FLOW SENSOR, one per engine.
2. Use ONLY approved 1/4 NPT fittings for transducer inlet and outlet, installer supplied.
3. Use appropriate fittings to match airplane fuel lines.
4. Use approved thread sealant. Torque pipe threads to a max of 15 ft. lbs. (180 in. lbs.)
5. INSIGHT supplied 1/3 turn bayonet mount power and signal harness P/N CAB-1000-809 by reducing turbulence in the rotor.
6. Mount on firewall or airframe with nomenclature facing up.
7. Installer may drill additional holes in bracket if necessary.
8. CONFIRM TRANSDUCER CASE IS GROUNDED TO AIRFRAME GROUND BEFORE POWERING UP
9. Confirm all wiring PRIOR to powering up
10. SEE INSTALLATION SCHEMATIC 1000-301 FOR TRUE FLOW SENSOR WIRING DETAILS.
11. SEE INSTALLATION SCHEMATIC 1000-816 FOR TRUE FLOW SENSOR WIRING DETAILS.
12. SPECIFY GPS MODEL WHEN ORDERING TRUE FLOW SENSOR, B = BENDIX KING, Z = APOLLO AND GARMIN.
13. FLOW SENSOR 30,000 PULSES PER GALLON UNLESS INDICATED BY -Knnmm.
14. FLOW SENSOR DATA SIGNAL 5V UNLESS -EPR, EXTERNAL PULL-UP RESISTOR REQUIRED, SEE WIRING DIAGRAM.

REVISIONS			
REV	DESCRIPTION	DATE	APPROVED
A	Added type B Windicator (1000-002-B).	June 15/00	JWT
B	ADDED ADDITIONAL MOUNTING OPTIONS.	July 10/01	R.R.



INSTALLATION

1. LOCATE WINDICATOR ON PANEL.
2. USE 1.375 ROUND PANEL PUNCH. INSIGHT P/N MCF-730BB-1-38.
3. CONNECT WINDICATOR CABLE TO WINDICATOR. CONNECTOR HAS A SNAPPING ACTION.
4. USING CIRCUMFERENCE OF WINDICATOR, PRESS FIRMLY INTO PANEL.
5. APPLY RTV TO BACK OF WINDICATOR, SUGGEST 1 DROP 4 PLACES.
6. HOLD IN PLACE UNTIL RTV IS CURED.
7. SEE INSTALLATION SCHEMATIC 1000-301 FOR TERMINATION DETAIL.

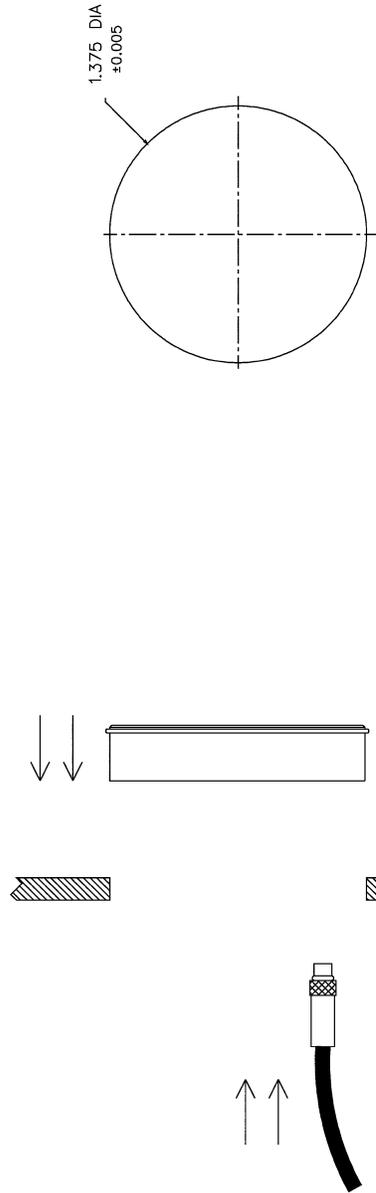
NOTES

1. FOR MOUNTING FRONT MOUNT WINDICATOR ONLY, P/N 1000-002-B
2. USE WINDICATOR CABLE P/N CAB-1000-507-06.
3. WHEN ORDERING THE WINDICATOR, MOUNTING MUST BE SPECIFIED.

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WINDICATOR REAR MOUNT TYPE 2	
Insight Instrument Corporation Box 122, Fort Erie, Ontario L2A 5M6	
FILE NAME: Windrrm2	DRAWING NO. 1000-305
DATE June 1/99	DRY PKD OGD REV B
SCALE none	SHEET 2 OF 3

REVISIONS		
REV	DESCRIPTION	DATE
A	Added type B Windicator (1000-002-B).	June 15/00 JMT
B	ADDED ADDITIONAL MOUNTING OPTIONS.	July 10/01 R.R.



RECOMMENDED
MOUNTING CUT-OUT
(NOT REQUIRED IF SURFACE MOUNTED)

INSTALLATION

1. LOCATE WINDICATOR ON PANEL.
2. USE 1.375 ROUND PANEL PUNCH. INSIGHT P/N MCF-730BB-1-38
3. PASS WINDICATOR CABLE FROM BEHIND PANEL THROUGH CUTOUT.
4. CONNECT WINDICATOR CABLE TO WINDICATOR. CONNECTOR HAS A SNAPPING ACTION.
5. USING CIRCUMFERENCE OF WINDICATOR, PRESS FIRMLY INTO PANEL.
6. APPLY RTV TO BACK OF WINDICATOR, SUGGEST 1 DROP 4 PLACES.
7. HOLD IN PLACE UNTIL RTV IS CURED.
8. SEE INSTALLATION SCHEMATIC 1000-301 FOR TERMINATION DETAIL.

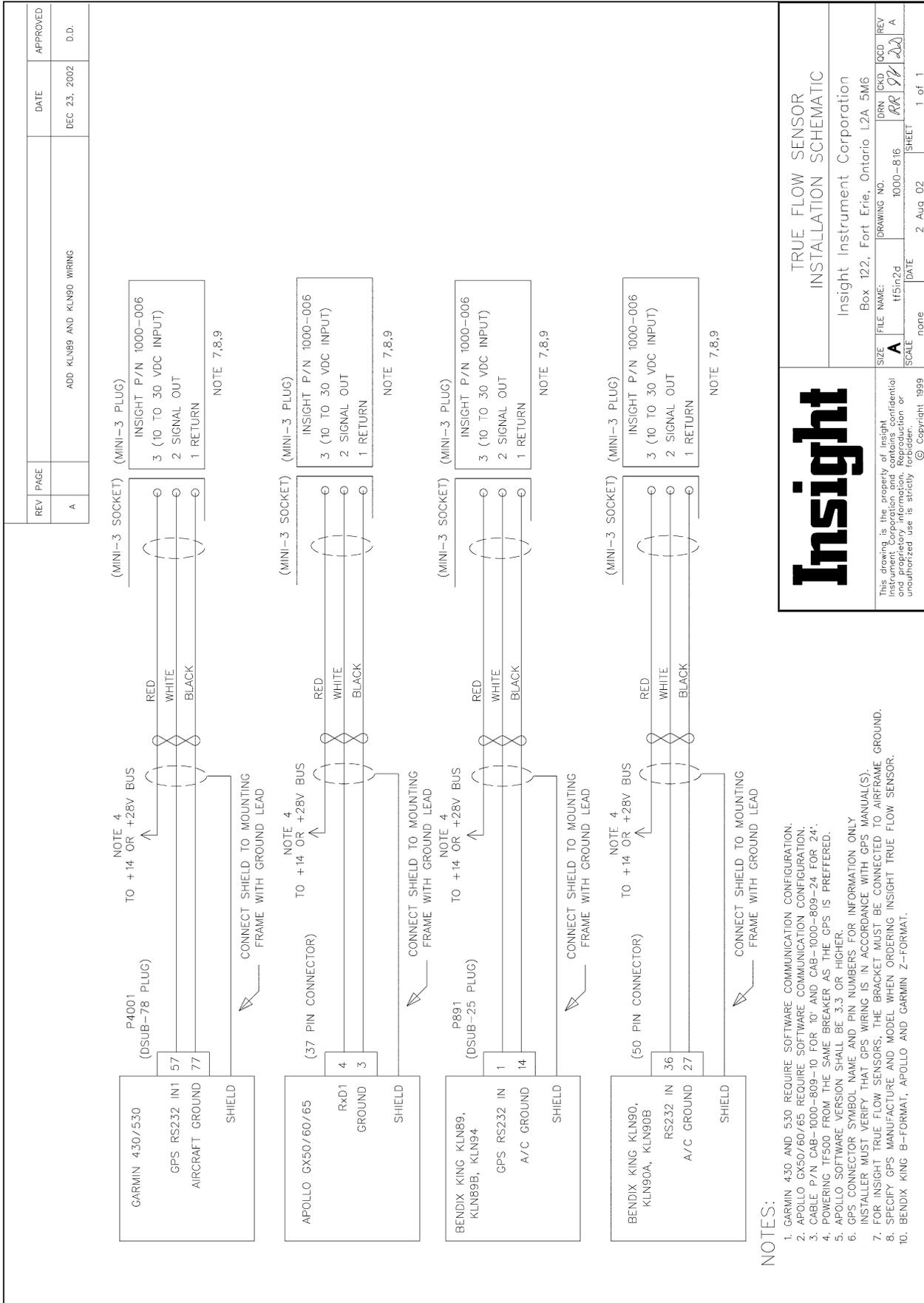
NOTES

1. FOR MOUNTING FRONT MOUNT WINDICATOR ONLY, P/N 1000-002-A
2. WINDICATOR CABLE P/N CAB-1000-507-06.
3. WHEN ORDERING THE WINDICATOR, MOUNTING MUST BE SPECIFIED.

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WINDICATOR FRONT MOUNT TYPE 3			
Insight Instrument Corporation			
Box 122, Fort Erie, Ontario L2A 5M6			
SIZE	DRAWING NO.	DRN	C/KG
Windfrm3	1000-305	Ins	Ins
SCALE	DATE	June 1/99	SHEET 3 OF 3
none			



- NOTES:**
- GARMIN 430 AND 530 REQUIRE SOFTWARE CONFIGURATION.
 - APOLLO GX50/60/65 REQUIRE SOFTWARE CONFIGURATION.
 - CABLE P/N CAB-1000-809-10 FOR 10' AND CAB-1000-809-24 FOR 24'.
 - POWERING IF500 FROM THE SAME BREAKER AS THE GPS IS PREFERRED.
 - APOLLO SOFTWARE VERSION SHALL BE 3.3 OR HIGHER.
 - GPS CONNECTOR SYMBOL NAME AND PIN NUMBERS FOR INFORMATION ONLY. INSTALLER MUST VERIFY THAT GPS WIRING IS IN ACCORDANCE WITH GPS MANUAL(S).
 - FOR INSIGHT TRUE FLOW SENSORS, THE BRACKET MUST BE CONNECTED TO AIRFRAME GROUND.
 - SPECIFY GPS MANUFACTURE AND MODEL WHEN ORDERING INSIGHT TRUE FLOW SENSOR.
 - BENDIX KING B-FORMAT, APOLLO AND GARMIN Z-FORMAT.

Insight

TRUE FLOW SENSOR
INSTALLATION SCHEMATIC

Insight Instrument Corporation
Box 122, Fort Erie, Ontario L2A 5M6

SIZE	FILE NAME	DRAWING NO.	DRN	ICD	DCD	REV
A	t45m2d	1000-816	AP	87	23	A
SCALE	DATE	SHEET				
none	2 Aug 02	1 of 1				

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